



CORONA CR
MISSION SUMMARY
AND
TELEMETRY ANALYSIS

MISSION 1114
AGENA 1660/PAYLOAD CR-14
MAY 1971

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GROUP 1

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1.0 SUMMARY

Mission 1114 utilized a THORAD booster (SLV-2H) S/N 538, Agena vehicle 1660, and payload system CR-14. The CR-14 payload system contained panoramic cameras S/N 328 and 329, and DISIC camera S/N 1-R. Payload profile and additional component serial numbers are included in Figure 7-1.

Lift-off occurred at 13:06:00.5 PST on 24 March 1971 from Vandenberg, SLC-3 west pad. All payload ascent events were normal with In-flight Reset (door ejection), A/P-to Orbit mode, instrumentation switchover, and panoramic camera transfer to orbit mode occurring as programmed. The orbit attained was within the three sigma of predicted.

The normal mission plan was 8/11 days with an actual of 7/9 days due to an excessive usage of vehicle control gas and an Agena "H" Timer failure on Rev. 246.

The performance of Panoramic cameras S/N 328 and 329 was normal throughout the flight. The panoramic film supply was not exhausted due to an Agena vehicle "H" Timer failure during Rev. 246. A total of 450 cycles was not recovered from the supply.

The panoramic camera A-to-B Transfer Sequence was performed on Rev. 104 and DISIC camera A-to-B Cut and Splice on Rev. 105. Both panoramic cameras and DISIC events were normal. The -1 mission recovery capsule was recovered by air catch on Rev. 115 at 1518 PST on 31 March 1971. The -2 mission recovery capsule was recovered by air catch on Rev. 260 at 1328 PST on 9 April 1971.

The clock system, command and instrumentation system, pressure make-up system, recovery systems, and the thermal environment were normal throughout the flight.

The early recovery of the -2 mission recovery capsule was necessitated due to the Agena vehicle "H" Timer failure on Rev. 246. The Life Boat recovery system timer was utilized for the -2 capsule recovery. All Life Boat systems performed satisfactorily.

On Rev. 159 [REDACTED] the first word loaded into the DSR was a 209 instead of a 201. The remaining thirty one (31) words loaded were correct. Post flight investigation of the problem revealed that the SILO decoder received an incomplete command message (SILO 311) because it was prematurely terminated due to a ground station problem. As a result the command was not executed and the SILO shift register in the Type 22 decoder was not cleared. The UNCLE Command system was then activated for DSR loading. During the first word of the 32 word DSR load a "One" bit from the SILO Type 22 decoder register was added to the DSR load from the Uncle Type 22 decoder register causing the first word of the DSR load to be a 209 instead of a 201. This was normal for the sequence of events that occurred during this command sequence.

2.0 SUBSYSTEM PERFORMANCE

2.1 Panoramic Cameras

Panoramic cameras S/N 328 and 329 performed normally during the -1 and -2 missions.

2.1.1 Film Consumption and Type

| | <u>FRAMES</u> | |
|---------------|----------------|----------------|
| | <u>Pan 328</u> | <u>Pan 329</u> |
| Sample | 20 | 41 |
| Pre-Launch | 119 | 120 |
| -1 Mission | 2988 | 2972 |
| -2 Mission | 2815 | 2800 |
| Left on Orbit | <u>220</u> | <u>229</u> |
| Total | 6162 | 6162 |

Film Supply Length and Type

| <u>Pan 328</u> | <u>Pan 329</u> |
|-----------------|-----------------|
| 7800 Ft/3414 | 7800 Ft/3414 |
| 1000 Ft/3404 | 1000 Ft/3404 |
| 7500 Ft/3414 | 7500 Ft/3414 |
| 16,300 Ft Total | 16,300 Ft Total |

2.2 DISIC Camera

The DISIC camera system performed normally throughout the -1 and -2 missions.

2.2.1 Film Consumption

| | <u>Frames</u> |
|---------------|----------------|
| | <u>Terrain</u> |
| Sample | 26 |
| Pre-Launch | 91 |
| -1 Mission | 2457 |
| -2 Mission | 2550 |
| Left on Orbit | <u>164</u> |
| Total | 5288 |

Length/Type

| <u>Terrain</u> | <u>Stellar</u> |
|----------------|----------------|
| 2200 Ft/3400 | 2000 Ft/3401 |

2.3 Command and Control

2.3.1 Command System

The real time command (RTC) system operation was satisfactory throughout the flight except for Rev. 159 [REDACTED]. On Rev. 159 [REDACTED] the SILO Command system malfunctioned during vehicle commanding making it necessary to load the DSR with the Uncle Command system. The first word loaded was a 209 instead of the desired 201. The balance of the load was correct. The DSR was then enabled prior to fade and was re-loaded correctly at 159 [REDACTED]. Post flight investigation of the problem revealed that the SILO decoder received an incomplete command message due to a ground station problem that caused a premature termination of the command. As a result, the last SILO command was not executed and the SILO shift register in the Type 22 decoder was not cleared. Therefore, when the DSR was loaded with the Uncle Command system the command words for the first word from both SILO and Uncle shift registers were added together resulting in the incorrect first word. This was normal for the Command Sequence that occurred on Rev. 159 [REDACTED]. Post event 2 testing was conducted to attempt to simulate a DSR problem without any malfunctions. (See Para. 5.0).

2.3.2 FMC Match

The Ramp to Orbit match was satisfactory throughout both missions. During 82.5% of the first mission operations and during 82.1% of the second mission operations, the mis-match error was less than 1.0%.

To compensate for the reduction of mission lifetime due to abnormal vehicle control gas usage, the orbit altitude was increased to minimize gas usage and hence extend mission life. The increase in altitude exceeded the pre-launch setting of the slope-programmer operating limits of 80.0 to 95.0 nautical miles. An improved ramp-to-orbit match would have been maintained if the operating limits had not been exceeded.

2.3.3 Exposure Control System

The slit width control programmer performed satisfactorily throughout the -1 and -2 missions.

2.4 Data System

2.4.1 Instrumentation

The instrumentation system performed satisfactorily throughout the -1 and -2 missions. Diagnostic data on Link Two (2) channel fifteen (15) appeared noisy after Rev. 43. Signal was "In and Out" for remainder of the mission. This anomaly was attributed to a problem in the Agena vehicle Link Two (2) signal strength.

2.4.2 Clock System

Mission 1114 payload clock performed satisfactorily throughout the flight. The correlation equation and constants are:

First Order Fit

$$\text{System Time} = A_0 + A_1 (\text{Clock Time})$$

$$A_0 = -0.2767048117484806 \text{ D } 05$$

$$A_1 = 0.9999997942584964 \text{ D } 00$$

$$\text{Sigma} = 0.01117422$$

Number of Points 257

Second Order Fit

$$\text{System Time} = A_0 + A_1 (\text{Clock Time}) + A_2 (\text{Clock Time})^2$$

$$A_0 = -0.2767052319235540 \text{ D } 05$$

$$A_1 = 0.9999999195136654 \text{ D } 00$$

$$A_2 = -0.7442426896944209 \text{ D } -13$$

$$\text{Sigma} = 0.00247612$$

Number of Points 257

2.4.3 SRV Tape Recorder

The SRV tape recorders for the -1 and -2 missions performed satisfactorily. A total of 210.2 minutes of data was recorded and processed from the two recorders (104.6 minutes for the -1 mission and 105.6 minutes for the -2 mission). Problems with the Automatic Data Processor delayed transmission of the customer data tape by two days.

2.5 Recovery

2.5.1 -1 Mission

The -1 recovery capsule was successfully recovered by air catch on Rev. 115 at 1518 PST on 31 March, 1971. All re-entry events were within tolerance with the impact approximately 10 miles North of the predicted. Refer to Table 7.5.

| | <u>Actual</u> | <u>Predicted</u> |
|-----------------|--------------------|--------------------|
| Impact Location | (24°11'N/162° 0'W) | (24°1'N/162° 14'W) |

2.5.2 -2 Mission

The -2 recovery capsule was successfully recovered by air catch on Rev. 260 at 1328 PST on 9 April, 1971. All re-entry events were within tolerance with the impact very near predicted. Refer to Table 7.5.

| | <u>Actual</u> | <u>Predicted</u> |
|-----------------|---------------------|--------------------|
| Impact Location | (17°58'N/153° 46'W) | (18°0'N/153° 58'W) |

3.0 ORBITAL PERFORMANCE

3.1 Orbital Parameters

| <u>Parameter</u> | <u>Predicted</u> | <u>Tolerance</u> | <u>Actual STC</u> | <u>Actual APF</u> |
|-----------------------------|---------------------|--|---------------------|---------------------|
| Period (Min.) | 88.67 | + .32, - .36 | 88.56 | 88.55 |
| Perigee (N.M.) | 84.5 | + 8, - 8 | 84.7 | 87.6 |
| Apogee (N.M.) | 145.9 | +12, -16 | 144.8 | 142.2 |
| Eccentricity | 88×10^{-4} | $+23 \times 10^{-4}, -30 \times 10^{-4}$ | 77×10^{-4} | 75×10^{-4} |
| Inclination (Deg.) | 81.50 | + .21, - .18 | 81.50 | 81.52 |
| Arg. of Perigee (Deg.) | 147 | +63, -57 | 130.6 | 129.0 |
| Regression Rate (Deg./Rev.) | 22.31 | - - | - | - |
| Perigee Latitude (Deg.) | 33 N | +59, -66 | - | - |

3.2 DMU Operation

Ground Track and period control were maintained during the flight by firing five (5) of the twelve (12) 3000 lb.-sec. DMU rockets. Refer to Table 3.2.1. The ground track and period control was affected by the change in orbit altitude to provide access to specific targets.

The ground track error at the ascending node ranged from 27.0 nautical miles east of nominal to 174.0 nautical miles west of nominal.

The firing of DMU rockets supported requirements due to the vehicle problem and due to the special access requested by the customer.

TABLE 3.2.1

DMU PERFORMANCE

| <u>Rocket No.</u> | <u>Rev. No.</u> | <u>System Time (Sec)</u> | <u>Period Change (Sec)</u> | <u>Velocity Change (Ft./Sec.)</u> | <u>Period at Firing (Min.)</u> | <u>Impulse (Lb./Sec.)</u> |
|-------------------|-----------------|--------------------------|----------------------------|-----------------------------------|--------------------------------|---------------------------|
| 1 | 27 | 47142 | 13.82 | 22.28 | 88.44 | 3072 |
| 2 | 48 | 4440 | 13.48 | 21.54 | 88.52 | 2974 |
| 3 | 134 | Stepped | Over | | | |
| 4 | 135 | 3972 | 15.35 | 24.55 | 88.31 | 3031 |
| 5 | 158 | 3957 | 15.45 | 24.71 | 88.43 | 3042 |
| 6 | 170 | 3927 | 15.40 | 24.60 | 88.63 | 3006 |

4.0 ENVIRONMENTAL CONTROL

4.1 Pressure Make-Up System

The pressure make-up system (PMU) operated properly throughout the flight. There were 109 panoramic camera operates for a total of 188.3 minutes which resulted in a gas consumption rate of 4.9 lbs/min of operate time. There were 262 DISIC camera operates for a total of 609.4 minutes for an alternate level gas consumption rate of 1.43 lbs/min of operate time.

4.2 Thermal Environment

The temperature data obtained during this flight indicated the temperature environment was within the pre-flight predictions for the duration of the flight. The averages of the panoramic camera temperatures ranged from 60°F to 62°F for S/N 328 and from 59°F to 64°F for S/N 329 during the -1 mission and 57°F to 64°F for S/N 328 and 59°F to 63°F for S/N 329 during the -2 mission. Refer to Tables 7.8.1 and 7.9.1 thru 7.9.4.

The on-orbit temperature profiles for Revs. 8, 105, and 186 are included in Figures 7.8.2 thru 7.8.14.

5.0 POST EVENT 2 TESTING

The Digital Storage Register (DSR) was loaded forty (40) times from various tracking stations in an attempt to repeat the anomaly that occurred on Rev. 159 [REDACTED]. No DSR loading malfunction occurred in any of the forty (40) loads.

6.0 HARDWARE DEFINITIONS

6.1 Agena

FTV 1660 was an Agena vehicle (SS-01B) and a THORAD Booster (SLV-2H)

S/N 538. The Agena was oriented nose first with the following configuration:

- 1) Twelve Thiokol DMU rockets. All Thiokol DMU rockets were 3000 lb-sec.
- 2) Three primary control gas spheres installed with -5 heavy control gas mixture.
- 3) -3 payload system with digital storage register (DSR) and capability of accepting both Silo and Uncle commands.
- 4) Ten panel, single wing, solar array system with two (2) 1H batteries. (Depleting system).
- 5) AFT payload-Doppler Beacon No. 5 and [REDACTED] Six Silo/Uncle commands for real time Doppler Beacon Control and Brush 18 (Doppler Beacon ON) and Brush 16 (Doppler Beacon OFF).
- 6) 3/4 speed Type VIII programmer (325 subcycles).
- 7) FTV 1660 was the second vehicle to utilize High Density Acid (oxidizer) and new fuel Hyperzine 300 (instead of IRFNA and UDMH).

6.2 Payload

The CR-14 payload configuration included the following:

1) Panoramic Camera

- a) Constant rotating type with a servo-controlled supply cassette.
- b) Digital Storage Register (DSR)/Cascade system utilized for camera enable/disable.
- c) Emergency program back-up available by RTC.

UHF 116/SILO 316 Emergency Program Select

UHF 118/SILO 318 Emergency Intermix Select

UHF 120/SILO 320 Emergency Mode Select

d) Exposure Control

1. Programmer control by SPC (51, 52, 17) and RTC
UHF 105/SILO 305.

2. Automatic slit width control. Override by RTC
UHF 101-126/SILO 301-326.

e) Filter Selection

1. Control by RTC UHF 103-104/SILO 303-304

2. The automatic filter change capability through the material change detector (MCD), was disconnected prior to launch.

f) Payload hardware modifications unique to this system.

1. The end of scan switch is now in series with the stow switch. Previously these switches were in parallel.

2. The instrument switch cluster has a 7-second timer added for back-up to the 20 second shutdown timer.

- 2) DISIC Camera
 - a) Mode select controlled by RTC UHF 124/SILO 324.
 - b) Both slave and independent modes of operation had a 1:1 ratio of stellar to terrain frames.
 - c) Operate off provided by RTC UHF 107/SILO 307.
- 3) FMC Programmer
 - a) Initiated by SPC 14 and SPC 27.
 - b) Control delay increment by RTC UHF 125/SILO 325.
 - c) Ramp profile provided by:
 - UHF 121/SILO 321 Eccentricity start level
 - UHF 122/SILO 322 Eccentricity half cycle level
- 4) Pressure Make-Up System
 - a) Enable/Disable controlled by RTC UHF 110/SILO 310.
 - b) Two (2) bottle system with dual range capability
 - c) PMU operation in low range with DISIC independent mode of operation.
- 5) Panoramic camera "A" to "B" Transfer available by RTC KIK-SILO 38.
- 6) DISIC camera "A" to "B" Transfer available by RTC KIK-SILO 39.
- 7) Yaw steering available by RTC UHF 106/SILO 306.
- 8) Agena tape recorder time shared with vehicle data.
- 9) SRV tape recorder available in -1 and -2 recovery capsules.
- 10) Payload weight: EWO = 1810 lbs.
- 11) Instrumentation: RTC UHF 127/SILO 327 operational-diagnostic data select.
- 12) Thermal configuration: The top black was reduced to 56 degrees on the fairing and 76 degrees on the barrel and conic.
- 13) Command system included a DSR for primary operation of the camera system with a two program/4 rev. intermix emergency capability.

~~TOP SECRET/C~~

HANDLE VIA [REDACTED]
CONTROL SYSTEM ONLY

6.3 Camera and Programmer Settings

6.3.1 Panoramic Cameras

| | <u>S/N 328</u> | <u>S/N 329</u> | | |
|---------------------|------------------|------------------|----------------|---------------|
| Filter Type | | | | |
| Primary | W/23, .037 Glass | W/25, .037 Glass | | |
| Alternate | W/23, .040 Glass | W/25, .040 Glass | | |
| Slit Width (Inches) | | | | |
| Position 1 | 0.115 | 0.144 | | |
| Position 2 | 0.145 | 0.177 | | |
| Position 3 | 0.178 | 0.218 | | |
| Position 4 | 0.283 | 0.308 | | |
| Failsafe | 0.155 | 0.190 | | |
| Auxiliary Optics | <u>Take-Up</u> | <u>Supply</u> | <u>Take-Up</u> | <u>Supply</u> |
| Filter | W-25 | W-25 | W-25 | W-25 |
| Aperture | F6.3 | F8.0 | F8.0 | F6.3 |

6.3.2 DISIC Camera

| | <u>Steller</u> | <u>Terrain</u> |
|--------------|----------------|----------------|
| Filter | None | W-12 |
| Aperture | F2.8 | F6.3 |
| Cycle Period | 9.375 | 9.375 |

6.3.3 Exposure Control Settings

| | <u>Seconds</u> |
|---|----------------|
| T-1 (20 second Increment) initial setting | 100 |
| T-2 DISIC Exposure to 1/500 | 320 |
| T-3 Slit Position 3 Duration | 40 |
| T-4 Slit Position 2 Duration | 240 |
| T-5 DISIC Exposure to 1/250 | 320 |
| T-6 (T6 Δ -T1) | 320 |

DISIC exposure time was constant at 1/500 second for this mission. No system function was controlled by T2 or T5.

Position parameters were used to facilitate T/M monitoring

6.3.4 FMC Control Settings

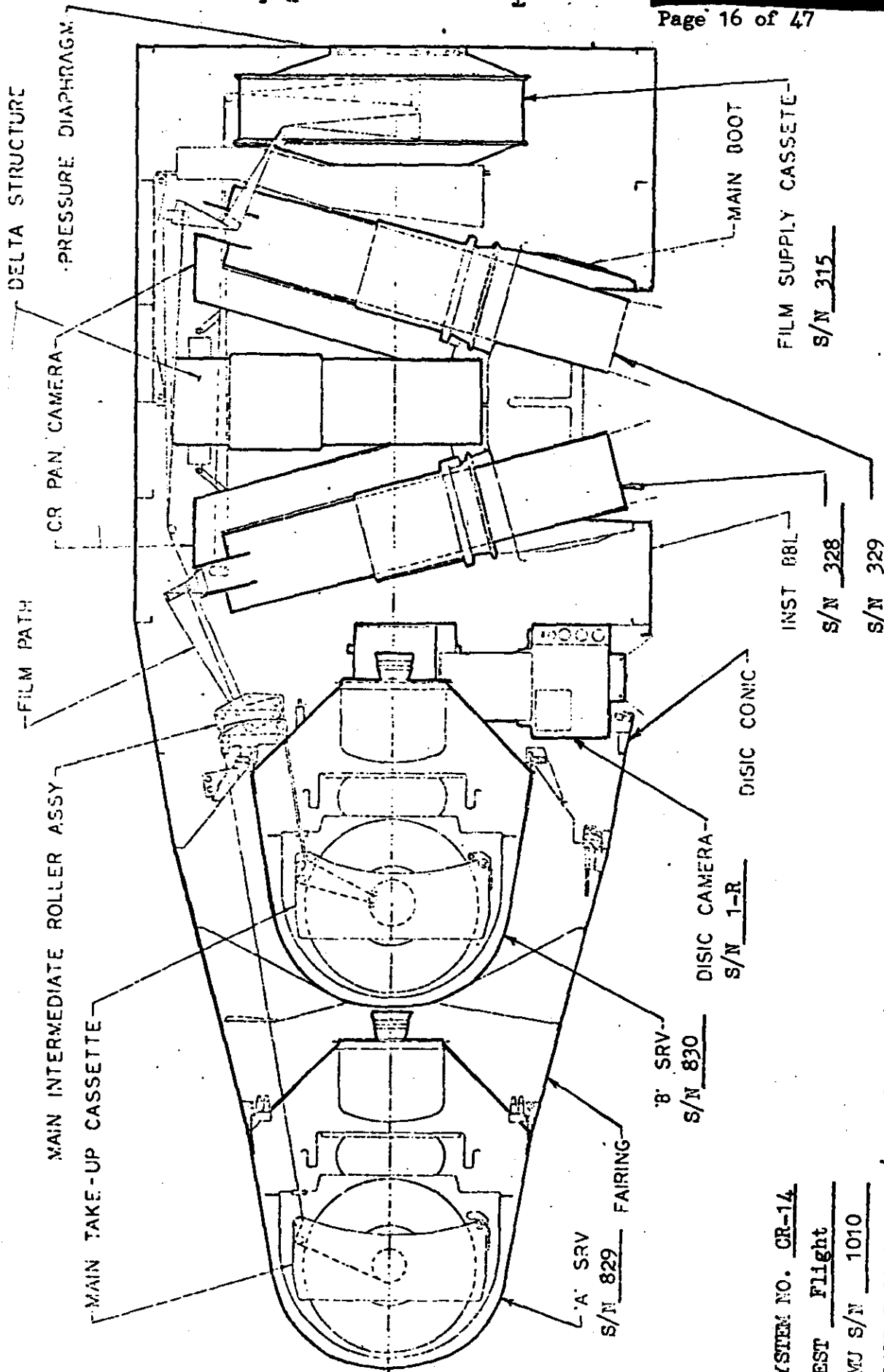
Eccentricity function

- 1) Period - 4188 seconds
- 2) Delay step increment - 50 seconds

Oblateness function

- 1) Period - 5248 seconds
- 2) Gain factor - 0.1136

2. PAYLOAD PROFILE AND SERIAL NUMBERS



SYSTEM NO. CR-14

TEST Flight

FMJ S/N 1010

SLOPE PROGRAMMER S/N 217

CLOCK S/N 629

SWITCH PROGRAMMER S/N 217

FIGURE 7.1

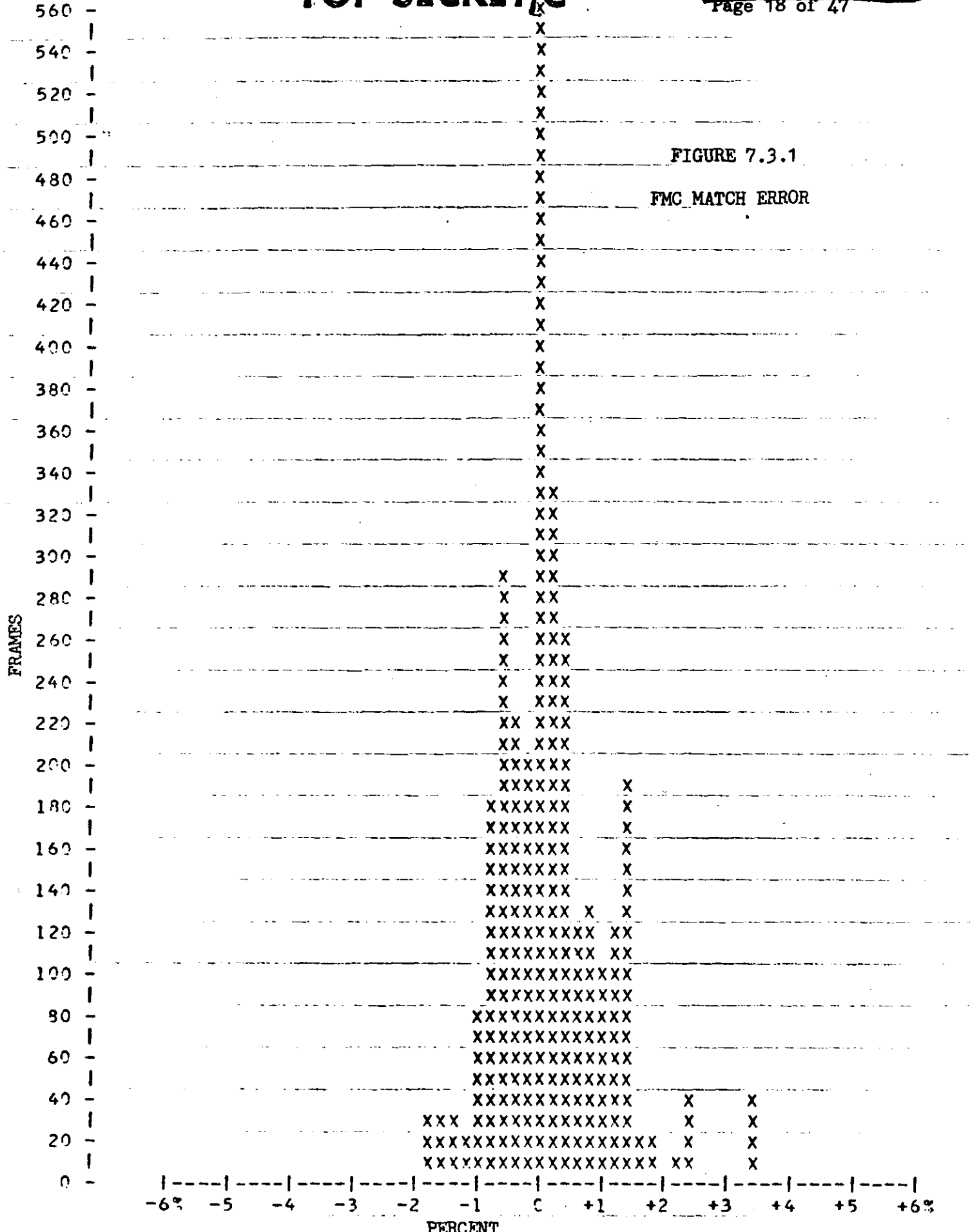
HANDLE VIA CONTROL SYSTEM ONLY

| REV | P | S | 1/2 POS. | SYSTEM CALIB. | OBL TUR | ECC TUR | I---- INST. 328 ----I | | | I---- INST. 329 ----I | | | 328/29 DIFF |
|-----|---|---|----------|---------------|---------|---------|-----------------------|-----------|-------------|-----------------------|-----------|-------------|-------------|
| | | | | | | | ACTUAL PERIOD | UNIT DEV. | SYSTEM DEV. | ACTUAL PERIOD | UNIT DEV. | SYSTEM DEV. | |
| 9 | 0 | 0 | 2 14 | 2.526 | 1845 | 743 | 2.512 | 0.57F | 0.57F | 2.527 | 0.03S | 0.03S | -0.6 |
| 32 | 0 | 0 | 1 13 | 1.852 | 3476 | 2301 | 1.835 | 0.91F | 0.93F | 1.840 | 0.68F | 0.66F | -0.2 |
| 48 | 0 | 0 | 1 15 | 1.834 | 3557 | 2438 | 1.822 | 0.61F | 0.64F | 1.831 | 0.17F | 0.15F | -0.4 |
| 97 | 0 | 0 | 4 4 | 2.075 | 3341 | 2424 | 2.058 | 0.79F | 0.80F | 2.062 | 0.62F | 0.61F | -0.1 |
| 129 | 0 | 0 | 5 8 | 2.008 | 3374 | 2538 | 1.995 | 0.65F | 0.67F | 2.000 | 0.43F | 0.42F | -0.2 |
| 161 | 0 | 0 | 2 1 | 2.084 | 3462 | 2020 | 2.075 | 0.42F | 0.43F | 2.083 | 0.06F | 0.04F | -0.2 |
| 226 | 0 | 0 | 3 3 | 2.080 | 3390 | 2388 | 2.080 | 0.01S | 0.01F | 2.080 | 0.02F | 0.01F | 0.0 |

NOTE: "F" = FAST and "S" = SLOW from the calibrated value

TABLE 7.2

PANORAMIC CAMERA CYCLE RATE ERRORS



MISSION 1114-1 AFT LOOKING--ORBIT MATCH
 MEAN= 0.16 ONE SIGMA= 0.84 TOTAL FRAMES=2938
 2470 FRAMES MATCHED ORBIT +/- 1% REPRESENTS 82.66% OF THE MISSION

FRAMES
560 -
540 -
520 -
500 -
480 -
460 -
440 -
420 -
400 -
380 -
360 -
340 -
320 -
300 -
280 -
260 -
240 -
220 -
200 -
180 -
160 -
140 -
120 -
100 -
80 -
60 -
40 -
20 -
0 -

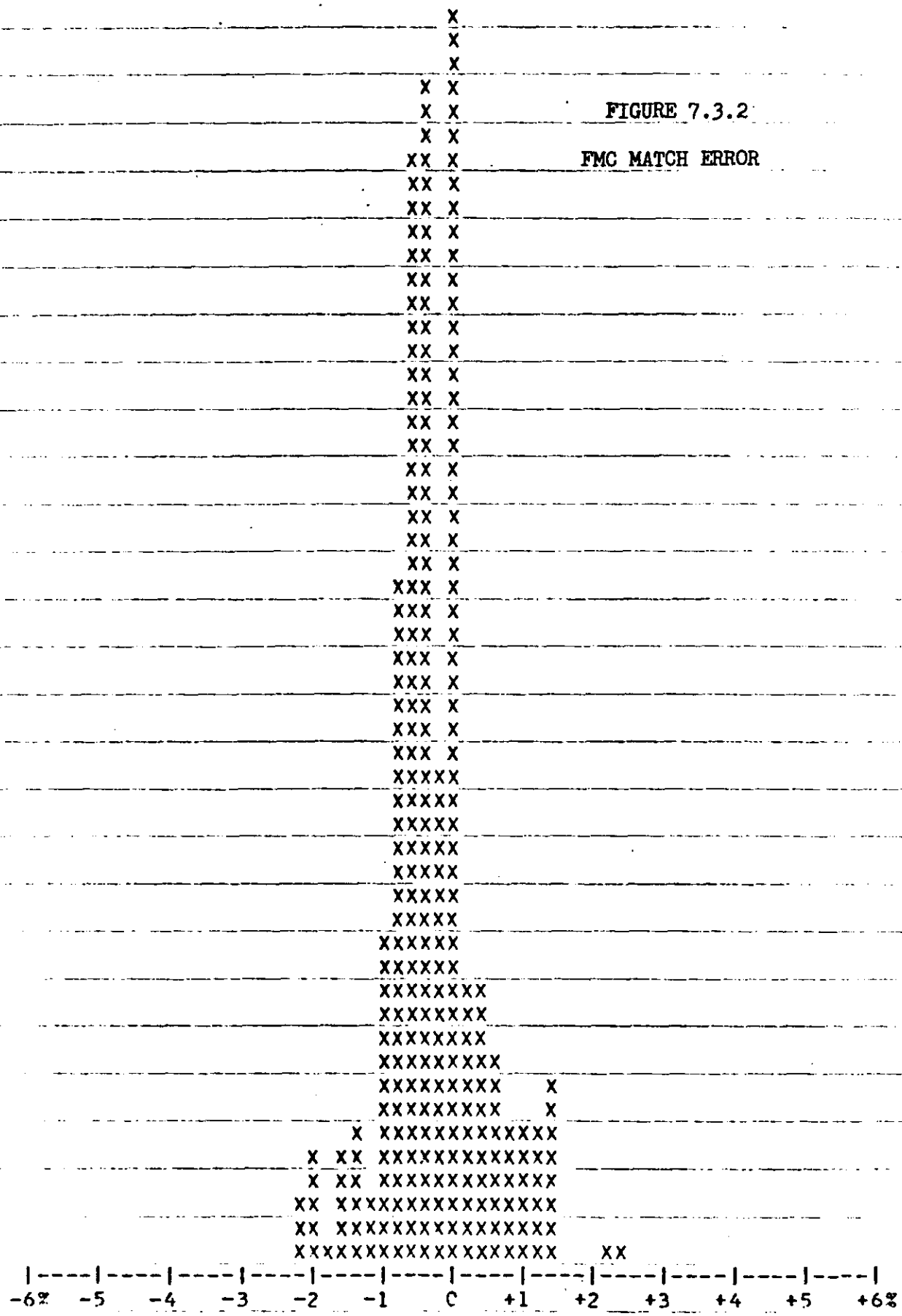


FIGURE 7.3.2

FMC MATCH ERROR

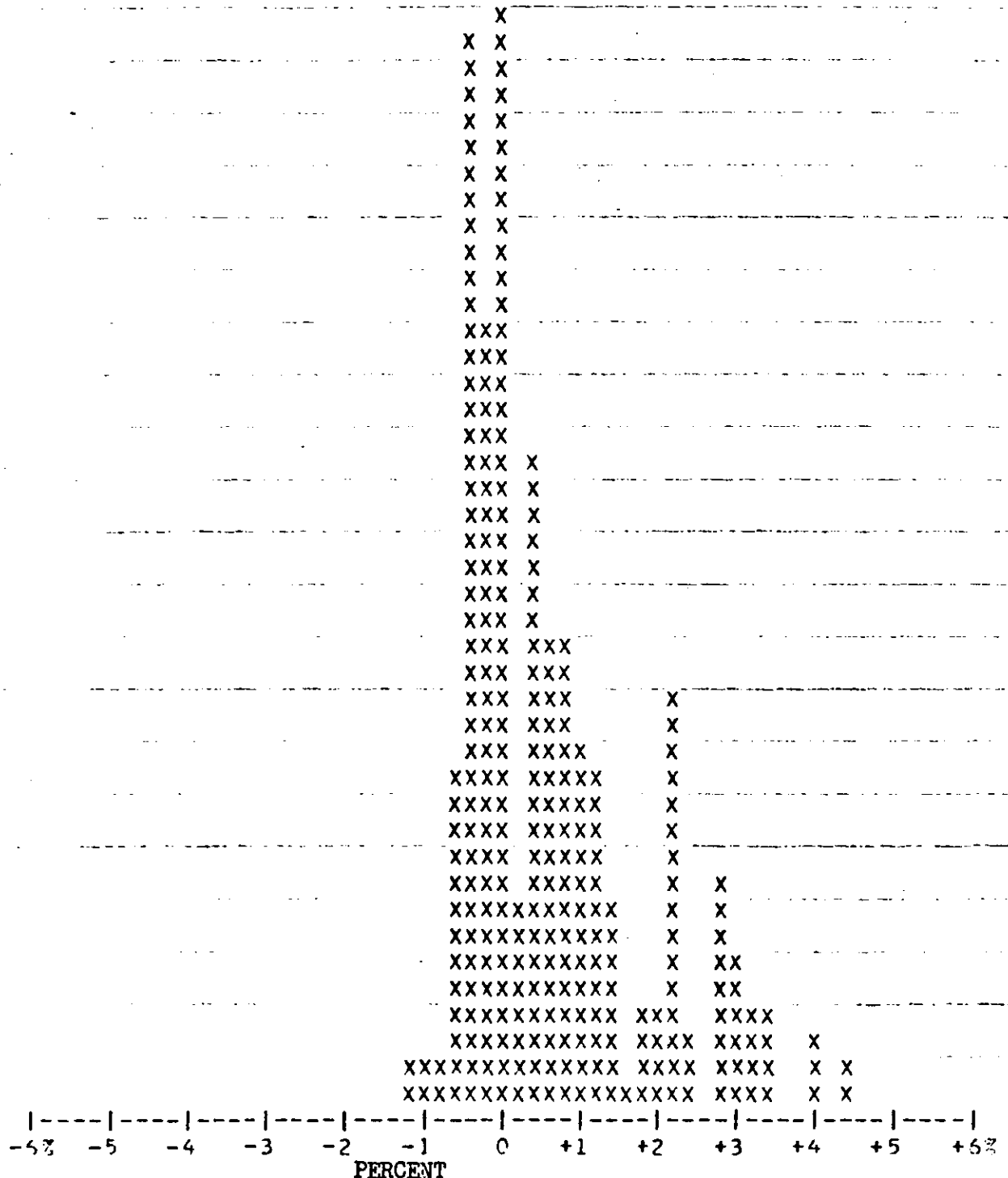
MISSION 1114-1 FWD LOOKING--ORBIT MATCH
 MEAN=-0.28 ONE SIGMA= 0.72 TOTAL FRAMES=2972
 2584 FRAMES MATCHED ORBIT +/- 1%, REPRESENTS 86.94% OF THE MISSION

FIGURE 7.3.3

FMC MATCH ERROR

FRAMES

500
540
520
500
480
460
440
420
400
380
360
340
320
300
280
260
240
220
200
180
160
140
120
100
80
60
40
20
0



MISSION 1114-2 AFT LOOKING--ORBIT MATCH
 MEAN= 0.65 ONE SIGMA= 1.16 TOTAL FRAMES=2918
 2123 FRAMES MATCHED ORBIT +/- 1%, REPRESENTS 72.76% OF THE MISSION

FRAMES

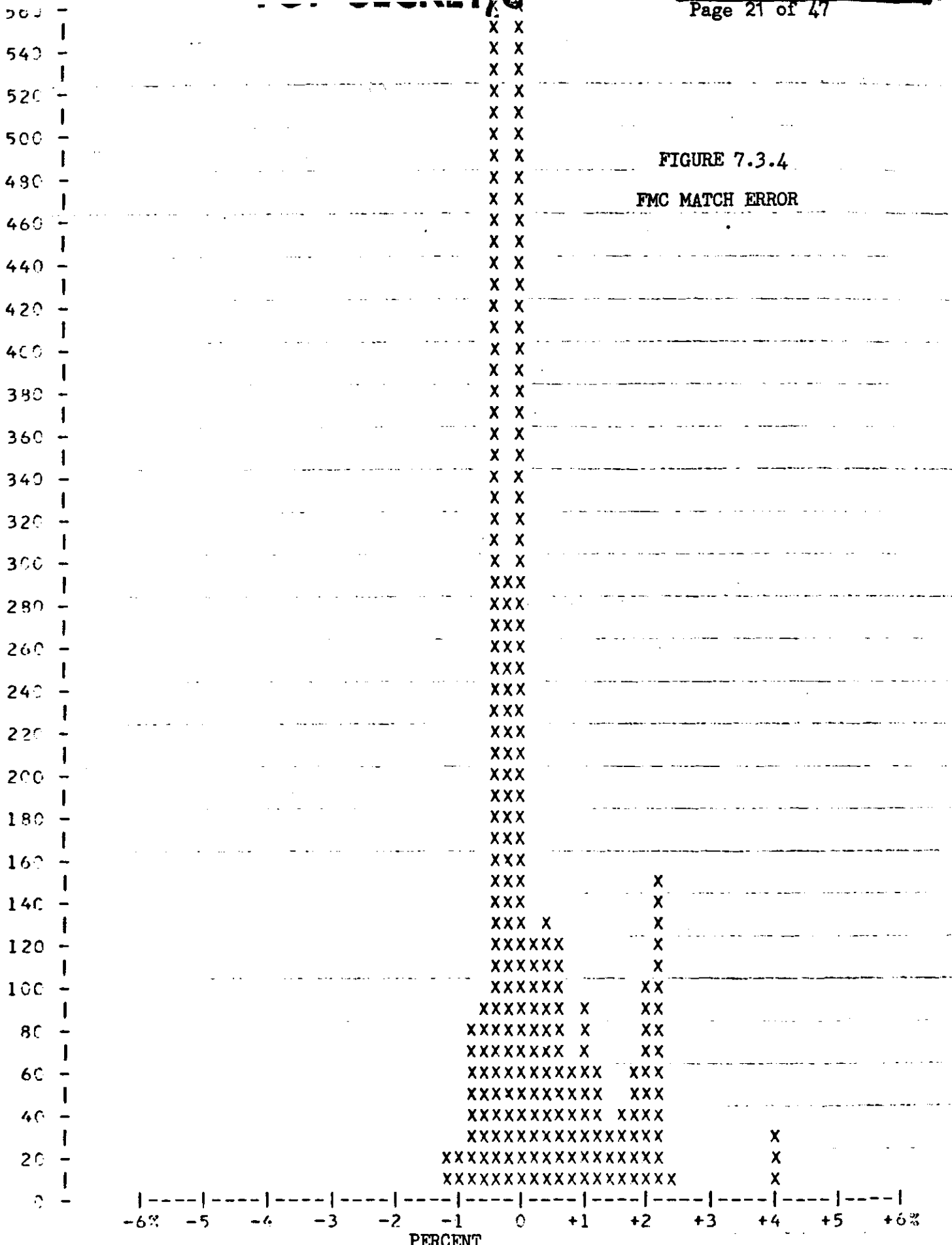


FIGURE 7.3.4

FMC MATCH ERROR

MISSION 1114-2 FWD LOOKING--ORBIT MATCH
 MEAN= 0.26 ONE SIGMA= 0.89 TOTAL FRAMES=2902
 2394 FRAMES MATCHED ORBIT +/- 1%, REPRESENTS 82.49% OF THE MISSION
 TOP SECRET
 HANDLE VIA [REDACTED]

MISSION 1114-2 AFT LOOKING, TOTAL FRAME COUNT- 2918

FRAME FREQUENCY DISTRIBUTION BETWEEN -6% AND +6% ORBIT MATCH

DISTRIBUTION OVER 61 POINTS INCREMENTED AT .2 PERCENT

PERCENT-FRAMES PERCENT-FRAMES

| | | | |
|------|-----|-----|-----|
| -0.2 | 296 | 0.0 | 417 |
| -0.4 | 405 | 0.2 | 80 |
| -0.6 | 134 | 0.4 | 251 |
| -0.8 | 24 | 0.6 | 182 |
| -1.0 | 24 | 0.8 | 175 |
| -1.2 | 24 | 1.0 | 135 |
| -1.4 | 0 | 1.2 | 133 |
| -1.6 | 0 | 1.4 | 79 |
| -1.8 | 0 | 1.6 | 11 |
| -2.0 | 0 | 1.8 | 40 |
| -2.2 | 0 | 2.0 | 39 |
| -2.4 | 0 | 2.2 | 160 |
| -2.6 | 0 | 2.4 | 33 |
| -2.8 | 0 | 2.6 | 0 |
| -3.0 | 0 | 2.8 | 94 |
| -3.2 | 0 | 3.0 | 60 |
| -3.4 | 0 | 3.2 | 37 |
| -3.6 | 0 | 3.4 | 36 |
| -3.8 | 0 | 3.6 | 0 |
| -4.0 | 0 | 3.8 | 0 |
| -4.2 | 0 | 4.0 | 25 |
| -4.4 | 0 | 4.2 | 0 |
| -4.6 | 0 | 4.4 | 24 |
| -4.8 | 0 | 4.6 | 0 |
| -5.0 | 0 | 4.8 | 0 |
| -5.2 | 0 | 5.0 | 0 |
| -5.4 | 0 | 5.2 | 0 |
| -5.6 | 0 | 5.4 | 0 |
| -5.8 | 0 | 5.6 | 0 |
| -6.0 | 0 | 5.8 | 0 |
| | | 6.0 | 0 |

TABLE 7.4.1

MISSION 1114-2 FWD LOOKING, TOTAL FRAME COUNT- 2902

FRAME FREQUENCY DISTRIBUTION BETWEEN -6% AND +6% ORBIT MATCH

DISTRIBUTION OVER 61 POINTS INCREMENTED AT .2 PERCENT

PERCENT-FRAMES PERCENT-FRAMES

| | | | |
|--|--|-----|-----|
| | | 0.0 | 731 |
| | | 0.2 | 118 |
| | | 0.4 | 127 |
| | | 0.6 | 119 |
| | | 0.8 | 55 |
| | | 1.0 | 93 |
| | | 1.2 | 61 |
| | | 1.4 | 33 |
| | | 1.6 | 39 |
| | | 1.8 | 63 |
| | | 2.0 | 101 |
| | | 2.2 | 148 |
| | | 2.4 | 14 |
| | | 2.6 | 0 |
| | | 2.8 | 0 |
| | | 3.0 | 0 |
| | | 3.2 | 0 |
| | | 3.4 | 0 |
| | | 3.6 | 0 |
| | | 3.8 | 0 |
| | | 4.0 | 25 |
| | | 4.2 | 0 |
| | | 4.4 | 0 |
| | | 4.6 | 0 |
| | | 4.8 | 0 |
| | | 5.0 | 0 |
| | | 5.2 | 0 |
| | | 5.4 | 0 |
| | | 5.6 | 0 |
| | | 5.8 | 0 |
| | | 6.0 | 0 |

TABLE 7.4.2

MISSION 1114-1 AFT LOOKING, TOTAL FRAME COUNT- 2988

FRAME FREQUENCY DISTRIBUTION BETWEEN -6% AND +6% ORBIT MATCH

DISTRIBUTION OVER 61 POINTS INCREMENTED AT .2 PERCENT

PERCENT-FRAMES PERCENT-FRAMES

| | | | |
|------|-----|-----|-----|
| | | 0.0 | 586 |
| -0.2 | 195 | 0.2 | 333 |
| -0.4 | 218 | 0.4 | 263 |
| -0.6 | 285 | 0.6 | 116 |
| -0.8 | 175 | 0.8 | 125 |
| -1.0 | 75 | 1.0 | 99 |
| -1.2 | 17 | 1.2 | 118 |
| -1.4 | 26 | 1.4 | 185 |
| -1.6 | 26 | 1.6 | 16 |
| -1.8 | 26 | 1.8 | 16 |
| -2.0 | 0 | 2.0 | 0 |
| -2.2 | 0 | 2.2 | 11 |
| -2.4 | 0 | 2.4 | 41 |
| -2.6 | 0 | 2.6 | 0 |
| -2.8 | 0 | 2.8 | 0 |
| -3.0 | 0 | 3.0 | 0 |
| -3.2 | 0 | 3.2 | 0 |
| -3.4 | 0 | 3.4 | 36 |
| -3.6 | 0 | 3.6 | 0 |
| -3.8 | 0 | 3.8 | 0 |
| -4.0 | 0 | 4.0 | 0 |
| -4.2 | 0 | 4.2 | 0 |
| -4.4 | 0 | 4.4 | 0 |
| -4.6 | 0 | 4.6 | 0 |
| -4.8 | 0 | 4.8 | 0 |
| -5.0 | 0 | 5.0 | 0 |
| -5.2 | 0 | 5.2 | 0 |
| -5.4 | 0 | 5.4 | 0 |
| -5.6 | 0 | 5.6 | 0 |
| -5.8 | 0 | 5.8 | 0 |
| -6.0 | 0 | 6.0 | 0 |

TABLE 7.4.3

~~TOP SECRET/C~~

HANDLE VIA 
CONTROL SYSTEM ONLY

MISSION 1114-1 FWD LOOKING, TOTAL FRAME COUNT- 2972

FRAME FREQUENCY DISTRIBUTION BETWEEN -6% AND +6% ORBIT MATCH

DISTRIBUTION OVER 61 POINTS INCREMENTED AT .2 PERCENT

PERCENT-FRAMES PERCENT-FRAMES

| | | | |
|------|-----|-----|-----|
| -0.2 | 210 | 0.0 | 532 |
| -0.4 | 503 | 0.2 | 121 |
| -0.6 | 469 | 0.4 | 115 |
| -0.8 | 288 | 0.6 | 91 |
| -1.0 | 137 | 0.8 | 60 |
| -1.2 | 31 | 1.0 | 58 |
| -1.4 | 55 | 1.2 | 57 |
| -1.6 | 54 | 1.4 | 75 |
| -1.8 | 13 | 1.6 | 0 |
| -2.0 | 48 | 1.8 | 0 |
| -2.2 | 34 | 2.0 | 0 |
| -2.4 | 0 | 2.2 | 11 |
| -2.6 | 0 | 2.4 | 10 |
| -2.8 | 0 | 2.6 | 0 |
| -3.0 | 0 | 2.8 | 0 |
| -3.2 | 0 | 3.0 | 0 |
| -3.4 | 0 | 3.2 | 0 |
| -3.6 | 0 | 3.4 | 0 |
| -3.8 | 0 | 3.6 | 0 |
| -4.0 | 0 | 3.8 | 0 |
| -4.2 | 0 | 4.0 | 0 |
| -4.4 | 0 | 4.2 | 0 |
| -4.6 | 0 | 4.4 | 0 |
| -4.8 | 0 | 4.6 | 0 |
| -5.0 | 0 | 4.8 | 0 |
| -5.2 | 0 | 5.0 | 0 |
| -5.4 | 0 | 5.2 | 0 |
| -5.6 | 0 | 5.4 | 0 |
| -5.8 | 0 | 5.6 | 0 |
| -6.0 | 0 | 5.8 | 0 |
| | | 6.0 | 0 |

TABLE 7.4.4

RE-ENTRY SEQUENCE OF EVENTS

| <u>Event</u> | <u>Delta Time (Seconds)</u> | | |
|---------------------------|-----------------------------|----------------|----------------|
| | <u>Nominal</u> | <u>Unit #1</u> | <u>Unit #2</u> |
| D-Timer Start | 0 | 0 | 0 |
| Arm | 6.0 ± .5 | 6.0 | 6.0 |
| Transfer | 81.0 ± .5 | 80.87 | 80.88 |
| Elec. Disconnect | 82.0 ± .5 | 81.83 | 81.88 |
| Separation | 83.0 ± .5 | 82.86 | 82.87 |
| Spin | 3.40 ± .30 | 3.27 | 3.41 |
| Retro | 7.55 ± .45 | 7.64 | 7.54 |
| Despin | 10.75 ± .54 | 10.83 | 10.64 |
| Thrust Cone Sep. | 1.50 ± .15 | 1.33 | 1.50 |
| "G" Switch Open | | | |
| Parachute Cover Ejec. | 26.0 ± 1.5 | 25.93 | 26.10 |
| Deceleration Chute Deploy | .58 ± .08 | .55 | .54 |
| Ablative Shell Disconn. | .58 ± .08 | .55 | .54 |
| Main Chute Bag Sep. | 10.25 ± 1.5 | 10.28 | 10.06 |
| Main Chute Deploy | .52 ± .13 | .54 | .56 |
| Main Chute Disreef | 4.50 ± .80 | 4.54 | 4.23 |
| K-10 Reset | 28.0 ± 1.9 | 27.85 | 28.21 |

TABLE 7.5

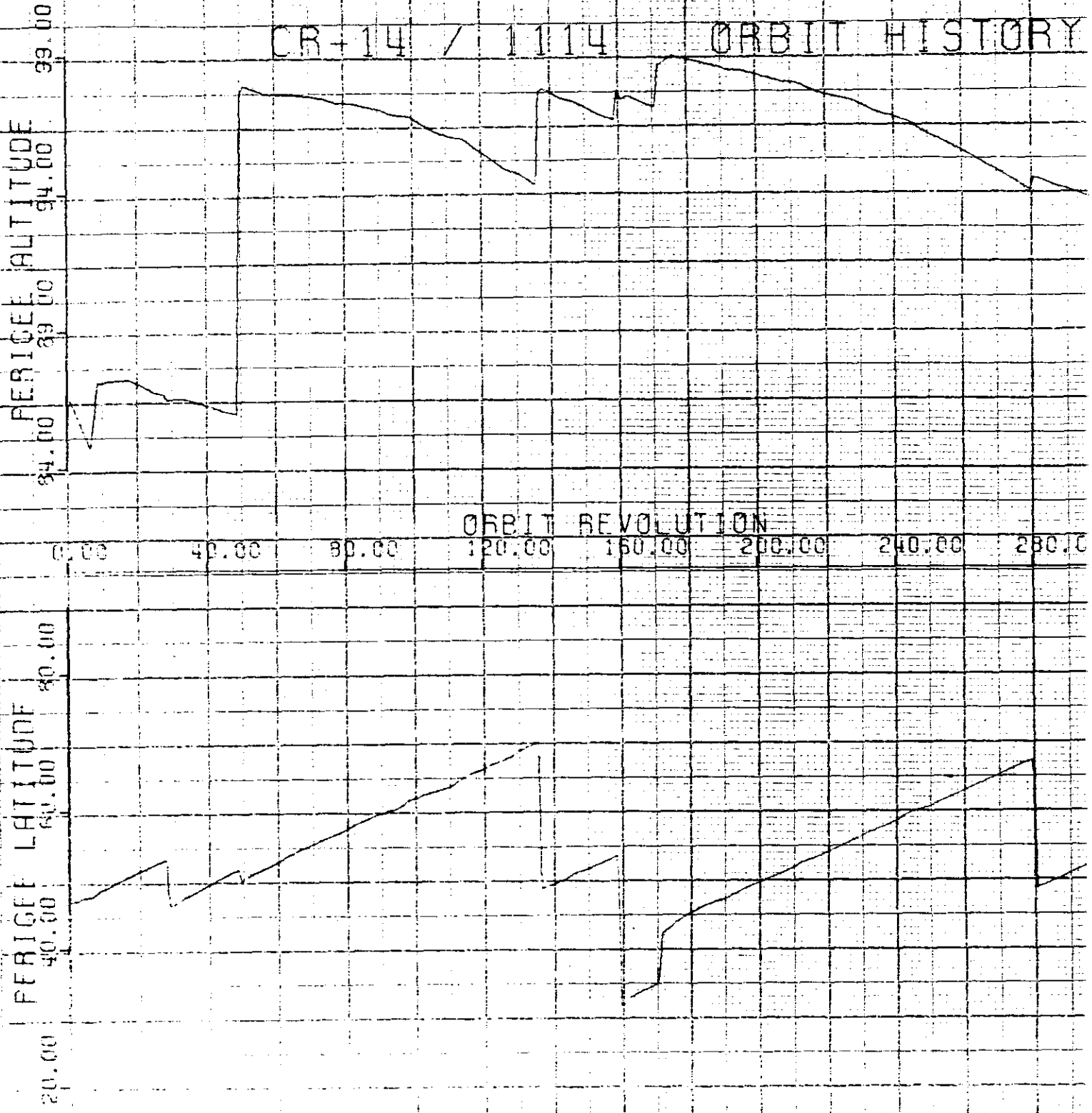


FIGURE 7.6.1

CR-14 / 1114 ORBIT HISTORY

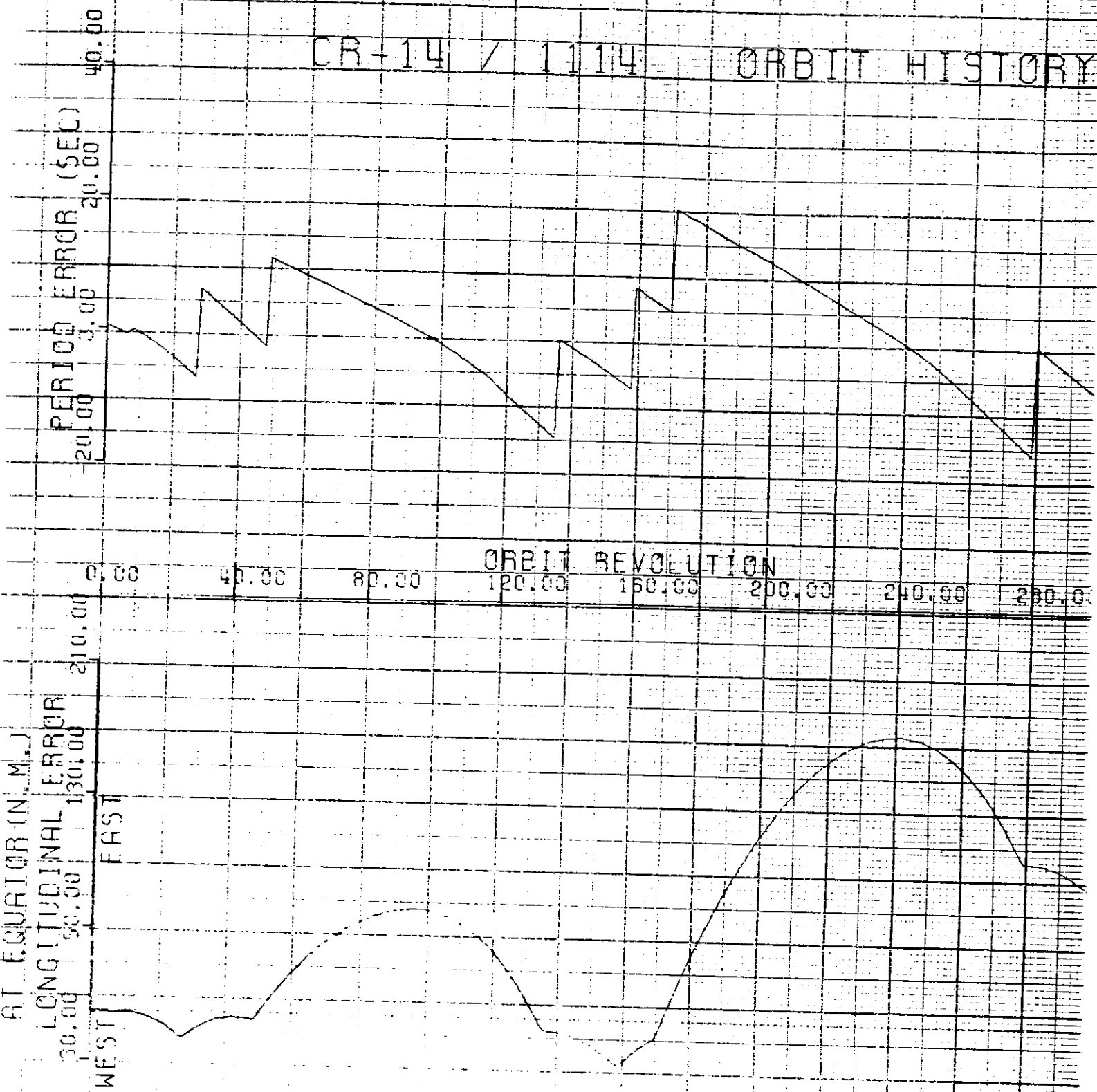


FIGURE 7.6.2

CR-14/1114/1660

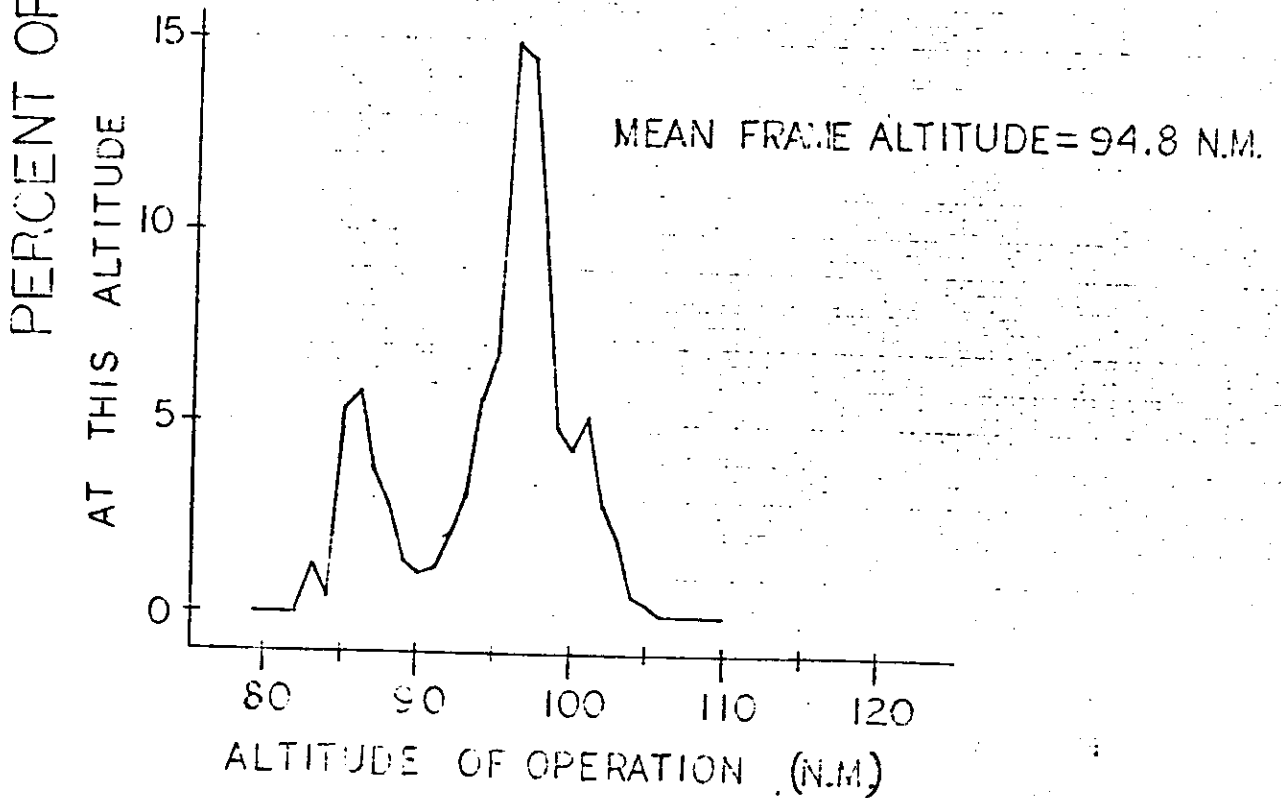
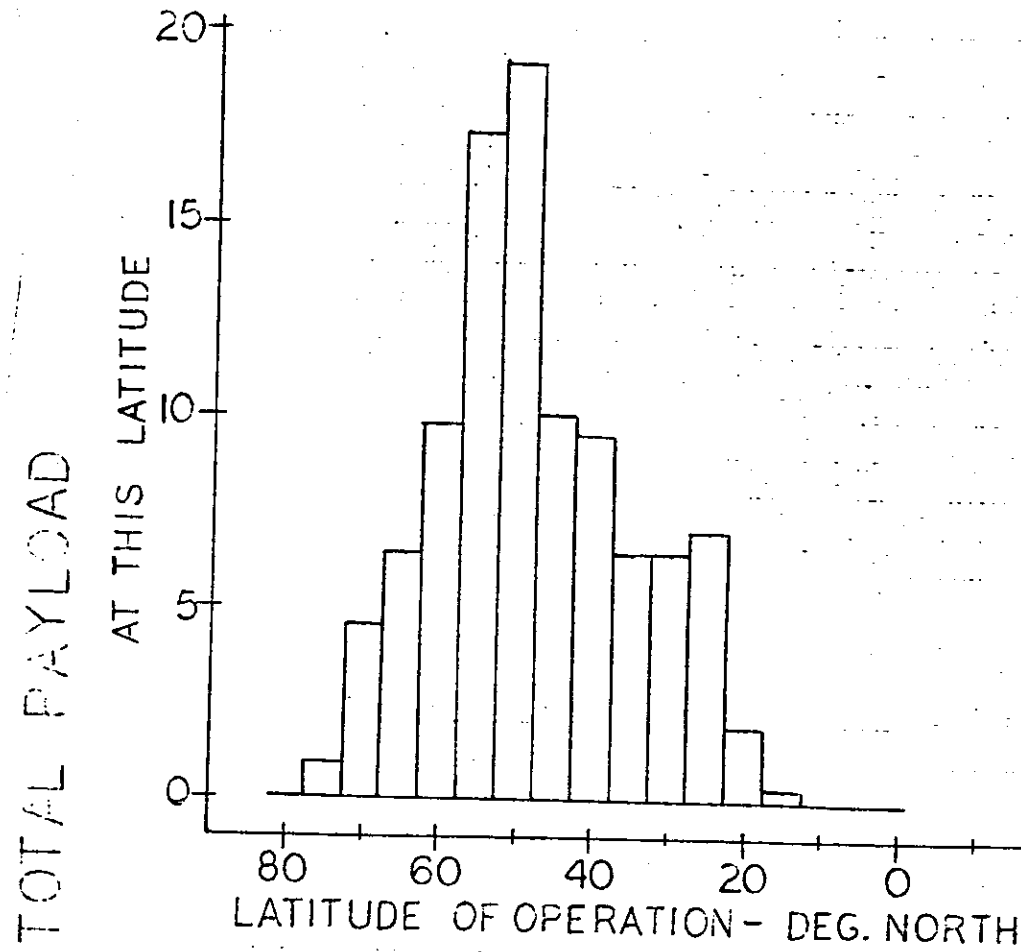


FIGURE 7.7

~~TOP SECRETIC~~

HANDLE VIA CONTROL SYSTEM

4-15-71

CR-14 FLIGHT VS PREDICTED TEMPERATURE

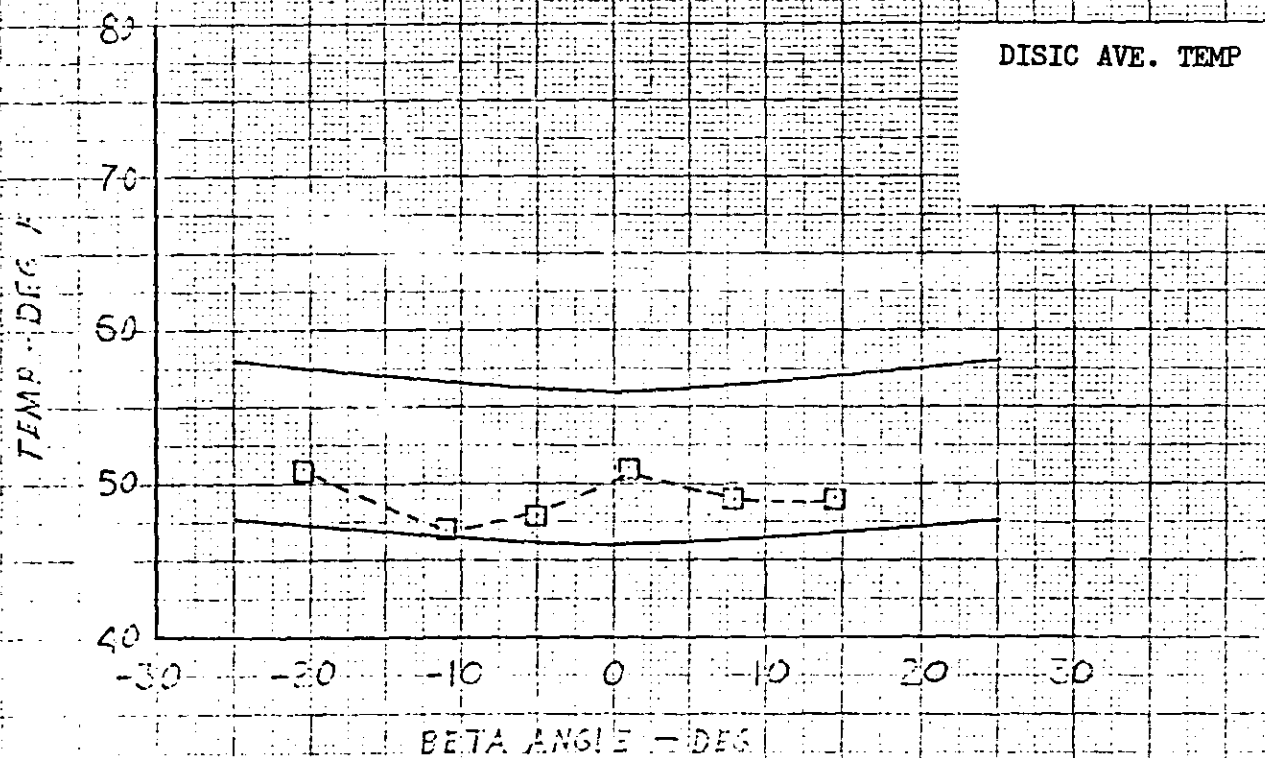
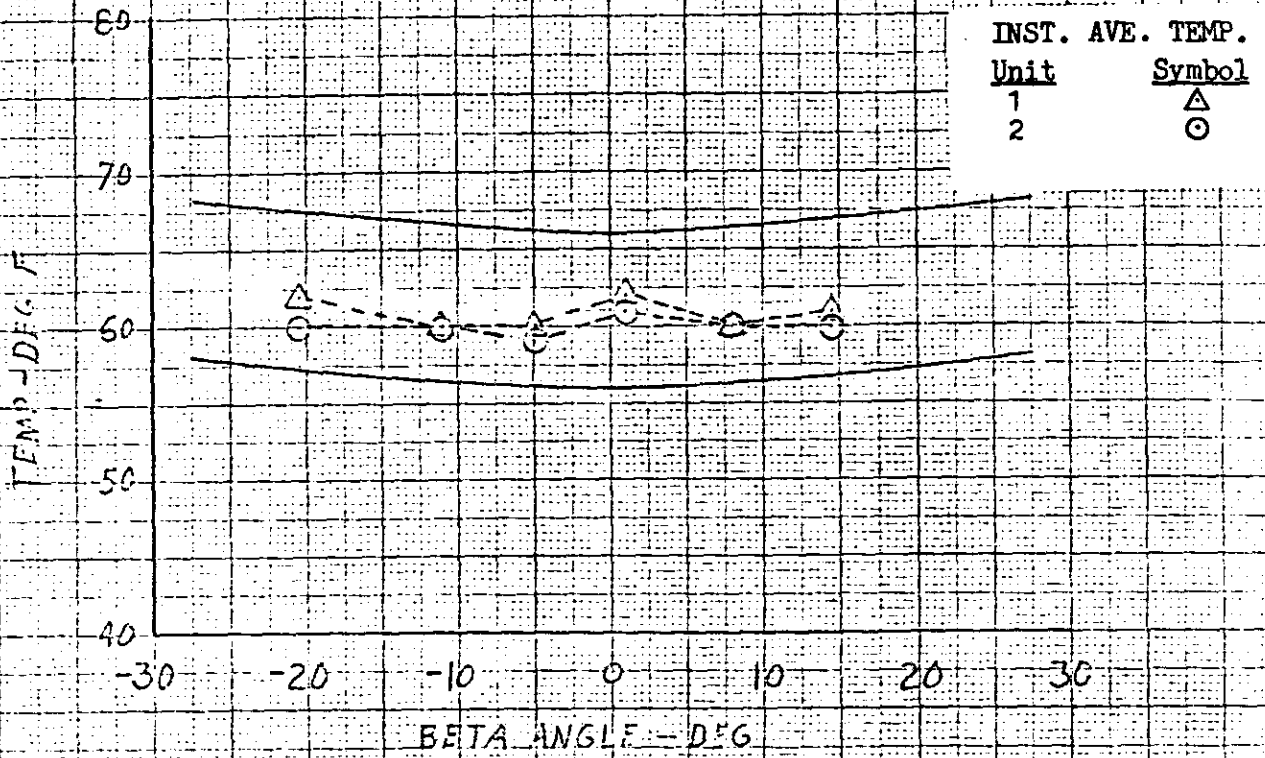


FIGURE 7.8.1

~~TOP SECRET~~

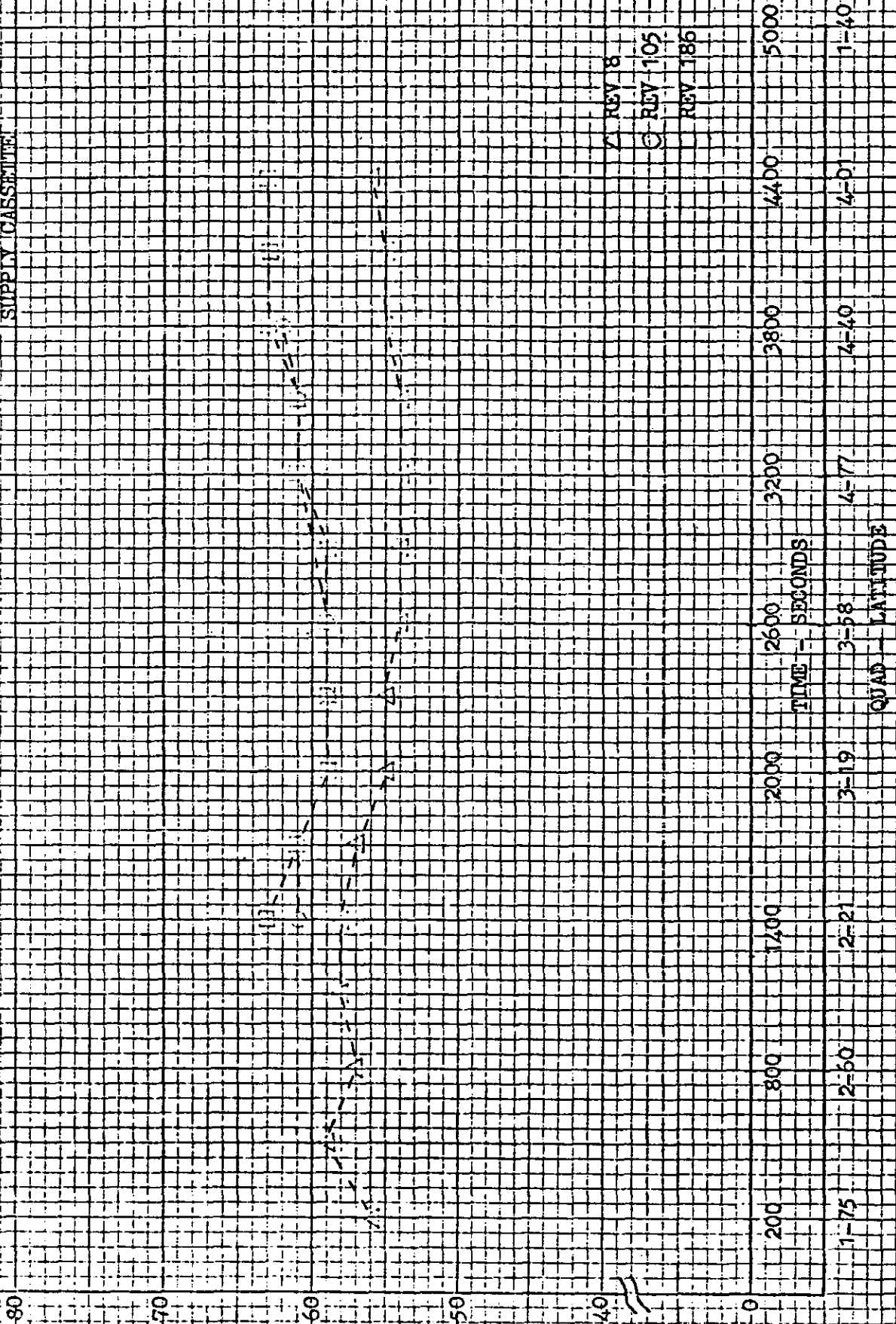
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MILLIMETER

CR-14

FLIGHT TEMPERATURE DATA
SUPPLY CASSETTE



ZI REV 18
© REV 105
REV 186

TOP SECRET

CONTROL SYSTEM ONLY

FIGURE 7.8.2

CR-17

FLIGHT TEMPERATURE DATA

PAN 1 GONE

80

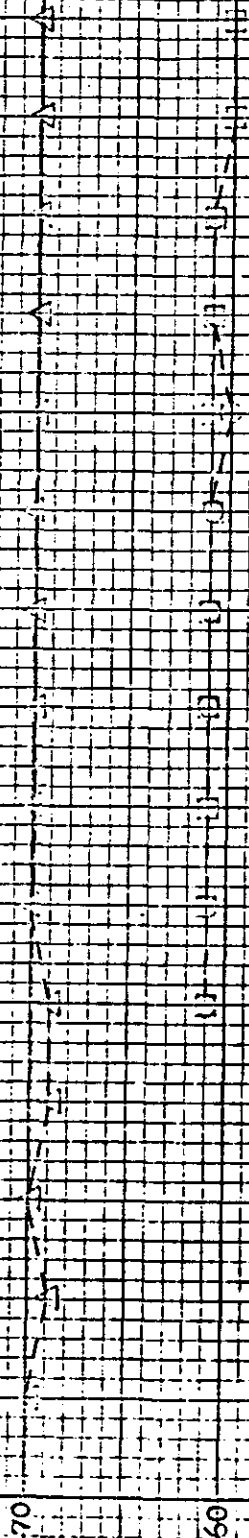
70

60

50

40

0



Δ REV 8
 ○ REV 105
 □ REV 186

| TIME - SECONDS | QUAD - LATITUDE |
|----------------|-----------------|
| 1-75 | 2-60 |
| 2-21 | 3-19 |
| 3-58 | 4-77 |
| 4-40 | 4-01 |
| 5-00 | 1-40 |

FIGURE 7.8.3

TEMPERATURE - DEGREES F

HANDLE VIA

CR-17

FLIGHT TEMPERATURE DATA

FAN CELL

80

70

60

50

40

200

800

1400

2000

2600

3200

3800

4400

5000

TIME - SECONDS

1-75

2-50

2-21

3-19

3-58

4-77

4-40

4-01

1-40

QUAD - LATITUDE

- △ REV 8
- REV 105
- REV 186

TEMPERATURE - DEGREES F

CONTROL SYSTEM

FIGURE 7.8.4