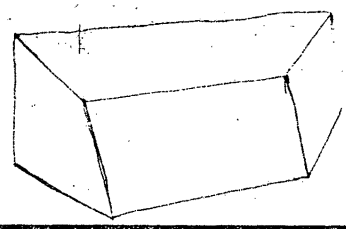
OPTION NUMBER AND DESCRIPTION	ADVANTAGES	DISADVANTAGES
OPTION # I = high sensitivty, horizon looking, minor lobe intercept system	1. DIRECTED against specific geographic area out of full POPPY Swath. 2. Locates Side-Lobe emissions. 3. Less Redundant data taken. 4. Higher accuracy location on long intercepts, 5. Has long term growth & extends Comb-Filter of 7107. 6. Large area can be searched.	1. Limited RF Range. 2. Sophisticated on-board Command and control. Must be synchronized in both cooperating spacecraft. 3. Reduces emitter scan info.
OPTION # II = MULTIPLE COLLECTION	1. Large amount of time & Area collection against priority job. 2. Relative simplicity. 3. Exploits the improved ground station processing capabilities. 4. Highest cost effectiveness, (Booster & Ground Station) 5. Inherent reliability thru redundancy. 6. can be launched launched by less expensive SCOUT Booster.	1. Production of large number of spacecraft. 2. Complex Spacecraft-separation control required.
OPTION # IV: HYBRID SYSTEM WITH BASIC DOWNWARD LOOKING SYSTEMS	1. Within its Frequency Range it: Locates Below Flight Line. Locates Brief Emissions. 2. Improved Accuracy through use in conjunction with 3. Locates Side Lobe emissions. 4. Provides less Redundant Data.	1. Very narrow RF Range (2.6-3.1GHz). 2. Small Instantaneous Search (1/10 of Main-beam System). 3. Requires Accurate Aspect system (½ degree) 4. Requires Large Fold-Out Antenna. 5. Sophisticated On-Board Processor 6. Accurate Location requires 3 cooperative spacecraft. 7. Requires Great ammount of Power 8. Due to cooperation between 3 separated spacecraft, common collection time rate is only 2/3 that normaly available.

OPTION # III
COMBINATION of
FEATURES of OPTION # II

TOP SECRET



TECHNICAL DATA

• Useful Orbital Life:	6 months
• Vehicle Weight: (including payload)	130 to 250 pounds
• Payload Weight:	to 200 pounds
• Dimensions:	approximately 35 x 35 x 13 inches (rocket motor nozzle extends an additional 4 inches along the thrust axis)
• Method of Stabilization:	spin stabilized (60 to 80 rpm)
	cold gas ejected through 2 spin nozzles
• Propulsion System:	any of 3 solid rocket motors for high and low orbit capability
Nominal Total Impulse:	
Model 0207:	1237 pound-seconds (513 pounds thrust for 2.41 seconds)
Model 1207:	17,650 pound-seconds (856 pounds thrust for 20.6 seconds)
Model 2207: (2 motors)	800 pound-seconds per motor (1000 pounds thrust for 0.8 second)
• Electrical Power System:	solar cells with secondary battery
	voltage limiter and control circuits for 22 to 29.25 volts dc
	power available/day 280 watt-hours
• Launch System:	orbital-launched from Agena or other space vehicle
	swing-out launching frame incorporating soft-spring system for satellite separation